



A Preliminary Study of Ice-Accretion Scaling for SLD Conditions

David N. Anderson
Ohio Aerospace Institute, Brook Park, Ohio

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David N. Anderson
Ohio Aerospace Institute, Brook Park, Ohio

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A PRELIMINARY STUDY OF ICE-ACCRETION SCALING FOR SLD CONDITIONS

David N. Anderson*
Ohio Aerospace Institute
Brook Park, Ohio 44142

Abstract

Proposed changes to aircraft icing certification rules are being considered by European, Canadian and American regulatory agencies to include operation in super-cooled large droplet conditions (SLD). This paper reports results of an experimental study in the NASA Glenn Icing Research Tunnel (IRT) to evaluate how well scaling methods developed for Appendix-C conditions might apply to SLD conditions. Until now, scaling studies have been confined to the FAA FAR-25 Appendix-C envelope of atmospheric cloud conditions. Tests were made in which it was attempted to scale to a droplet MVD of 50 μm from clouds having droplet MVD s of 175, 120, 100 and 70 μm . Scaling was based on the Ruff method with scale velocities found either by maintaining constant Weber number or by using the average of the velocities obtained by maintaining constant Weber number and constant Reynolds number. Models were unswept NACA 0012 wing sections. The reference model had a chord of 91.4 cm. Scale models had chords of 91.4, 80.0 and 53.3 cm. Tests were conducted with reference airspeeds of 100 and 150 kt (52 and 77 m/s) and with freezing fractions of 1.0, 0.6 and 0.3. It was demonstrated that the scaled 50- μm cloud simulated well the non-dimensional ice shapes accreted in clouds with MVD 's of 120 μm or less.

Nomenclature

A_c	Accumulation parameter, dimensionless
b	Relative heat factor, dimensionless
c	Airfoil chord, cm
d	Cylinder diameter or twice the leading-edge radius of airfoil, cm
K	Splashing factor, defined in Introduction, dimensionless
K_0	Modified inertia parameter, dimensionless
LWC	Cloud liquid-water content, g/m^3
MVD	Water droplet median volume diameter, μm
n	Freezing fraction, dimensionless

*Senior Research Associate, Member

Oh	Ohnesorge number, defined in Introduction, dimensionless
Re	Reynolds number of model, dimensionless
Re_δ	Reynolds number of water droplet, dimensionless
t_{st}	Static temperature, $^{\circ}\text{C}$
t_{tot}	Total temperature, $^{\circ}\text{C}$
V	Air velocity, m/s
We	Weber number based on droplet size and water properties, dimensionless
β_0	Collection efficiency at stagnation line, dimensionless
θ	Air energy transfer parameter, $^{\circ}\text{C}$
τ	Accretion time, min
ϕ	Droplet energy transfer parameter, $^{\circ}\text{C}$

Subscripts

R	Reference
S	Scale

Introduction

Proposed new icing certification rules to include testing with super-cooled large droplet (SLD) in addition to Appendix-C conditions are being considered for implementation within the next few years. When these new rules are established, test facilities will need to provide means for testing at or simulating SLD conditions. Existing Appendix-C calibrations will need to be expanded to include the new conditions. In addition, scaling techniques will be needed both to permit SLD-cloud testing of models smaller than full size and to simulate SLD droplet size effects with tests using Appendix-C MVD s. This report describes tests made in March 2001 in the IRT to provide a preliminary study of how effectively existing scaling methods can be applied to scale SLD drop sizes to Appendix-C conditions.

For some time it has been recognized that droplet splashing and breakup may influence ice accretion.¹ Studies have identified parameters with which droplet splashing effects can be correlated.^{2,3} Reference 3 introduced the K factor, $K = Oh Re_\delta^{1.25}$, where $Oh = We^{1/2}/Re_\delta$ is the Ohnesorge number. In these relationships, both We and Re_δ are based on the droplet

MVD and water properties. The K factor increases with velocity and *MVD*, and splashing only occurs when K is greater than a threshold value. Typically, K will be greater for SLD drop sizes than for those within Appendix C. Whether this K factor or its splashing threshold applies in icing conditions is not known, but it is thought that splashing in SLD conditions could have a much greater effect on ice accretion than for Appendix-C clouds. If so, scaling for SLD conditions may require new approaches.

For scaling within Appendix C, the Ruff scaling method⁴ has been shown to be effective. As used in atmospheric-pressure tunnels, this method maintains constant values of modified inertia parameter, K_0 , accumulation parameter, A_c , and two energy-balance terms: the freezing fraction, n , and either the water-energy transfer parameter, ϕ , or the air-energy-transfer parameter, θ . The stagnation collection efficiency, β_0 , is a function only of K_0 , so that matching K_0 assures that β_0 for the scale and reference test will be the same. With $\beta_{0S} = \beta_{0R}$, matching A_c assures that the quantity of ice accreted, relative to the chord, for the scale and reference test will be identical. The freezing fraction, n , is a heat-balance parameter with a strong effect on the shape of the ice, and ϕ and θ also have a small effect on ice shape. n , ϕ , and θ come from the heat-balance analysis of Messinger.⁵ These requirements produce four equations that can be solved simultaneously to determine the scale droplet size, spray time, liquid-water content and temperature.

In the Ruff method, the user chooses the scale velocity. In addition to other approaches, the scale velocity can be determined by requiring that either We or Re be matched between scale and reference conditions. Scaling studies in the IRT^{6, 7} showed that scale tests produced the closest match to reference ice shapes when a scale velocity was used that was the average of that found from keeping We constant and that from keeping Re constant between scale and reference cases.

The study of reference 1 showed that splashing depends in part on the thickness of the water film on the surface. Therefore, because the liquid water-film thickness must be dependent on the freezing fraction, different SLD effects may be evident for different freezing fractions. The present study was performed with freezing fractions of 0.3, 0.6 and 1.0. Rime tests were included because they provide a good check on tunnel calibrations. If the tunnel calibration is consistent, the correct scale rime accre-

tion is always obtained as long as the modified inertia parameter, K_0 , and accumulation parameter, A_c , are matched between scale and reference.

There have been no previous experimental studies of scaling for SLD conditions. However, a preliminary analysis⁸ of scaling from 200 μm to Appendix-C drop sizes of 40 or 50 μm concluded that to satisfy $K_{0S} = K_{0R}$, as the Ruff method requires, scale model size would need to be 1/5th the reference or smaller. Scaling for models smaller than 1/3rd the reference have not been successfully demonstrated, so there would be little confidence in the results of scaling to 1/5th size. Consequently, the study concluded that there were presently no validated ways to simulate SLD tests by using Appendix-C droplet sizes. However, a later study⁹ demonstrated that the requirement of equating scale and reference values of K_0 could be relaxed providing the stagnation collection efficiency, β_0 , matched within about 10%. For the present study, this relaxed standard was used, and scale models from 1/1.7th to 1/1 of the reference size were used.

The reference conditions for this series of tests used four of the five specific *MVD-LWC* combinations for which the IRT has been calibrated in the SLD regime.¹⁰ The calibration was made for 3 airspeeds: 100, 150 and 200 kt. For the tests reported here, only the 100- and 150-kt reference airspeeds were used. Table I shows the conditions chosen for each reference case. The cases were assigned a 3-digit number followed by 'SLD'. The first digit relates to the reference airspeed, with 1 indicating 100 kt and 2, 150 kt. The second digit refers to the reference *MVD*, with 2 representing 175 μm , 3, 120 μm , 4, 100 μm and 5, 70 μm . The third digit in the case designation pertains to the freezing fraction, with 0 used for $n = 1.0$, 2 for $n = 0.6$ and 5 for $n = 0.3$. The reference model for all tests was a 91.4-cm-chord NACA 0012 full-span wing section.

Test Description

The tests were performed in the NASA Glenn Icing Research Tunnel (IRT) in March 2001. The IRT is a closed-loop refrigerated icing tunnel. In the summer of 2000 a complete cloud calibration¹⁰ was completed following the installation of a new heat exchanger and other upgrades.¹¹ The tunnel loop is shown in figure 1(a). As part of the modifications, new C- and D- corner turning vanes were installed and the tunnel was extended to the west by 6 m to provide room for the new heat exchanger. This extension on the tunnel shell created more distance

Table I. Reference Conditions for SLD Scaling
Reference Model: 91.4-cm-Chord NACA 0012

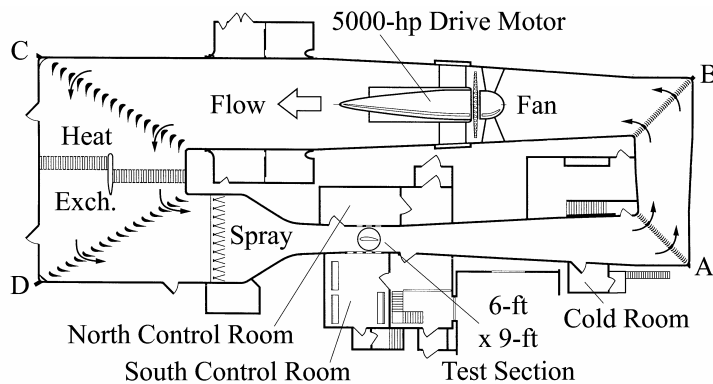
Case	t_{sts} , °C	t_{tot} , °C	V , kt	V , m/s	MVD , μm	LWC , g/m ³	τ , min	β_0 , %	A_c	n
130SLD	-29	-28	100	51.5	120	1.01	16.1	94	1.90	1.00
122SLD	-23	-22	100	51.5	175	1.68	9.7	96	1.90	0.60
132SLD	-17	-15	100	51.5	120	1.01	16.1	94	1.90	0.60
230SLD	-27	-24	150	77.2	120	0.69	15.8	95	1.90	1.00
250SLD	-23	-20	150	77.2	70	0.60	18.1	90	1.90	1.00
222SLD	-20	-17	150	77.2	175	0.99	11.0	96	1.90	0.60
232SLD	-16	-13	150	77.2	120	0.69	15.8	94	1.90	0.60
242SLD	-16	-13	150	77.2	100	0.68	16.0	93	1.90	0.60
252SLD	-14	-11	150	77.2	70	0.60	18.1	90	1.90	0.60
225SLD	-10	-7	150	77.2	175	0.99	11.0	96	1.90	0.30
235SLD	-8	-5	150	77.2	120	0.69	15.8	94	1.90	0.30
245SLD	-8	-5	150	77.2	100	0.68	16.0	93	1.90	0.30

from the turning vanes to the spray bars on the south side of the settling chamber. The IRT has 10 spray bars, an arrangement in use since 1998.

The IRT spray-bar system uses two separate water manifolds. The water pressure of each can be controlled independently. This set-up allows both Standard and Mod-1 nozzle sets to be installed at all times so that it's possible to switch rapidly and easily between nozzle sets as test conditions require. All results presented here were obtained with Mod-1 nozzles.

The models used for these tests were all NACA 0012 wing sections. They included a full-span (183-cm) 91.4-cm-chord fiberglass model pictured in figure 1(b) and 61-cm-span aluminum airfoil sec-

tions with 80.0- and 53.3-cm chord. The 61-cm-span models were mounted between splitter plates as shown typically in figure 1(c). At the bottom, the splitter plates were supported on a stand mounted to the tunnel turntable. The upper splitter plates were secured to a frame attached to a hollow cylinder at the ceiling of the tunnel. This cylinder fit closely around a vertical pipe section attached to the tunnel ceiling. This arrangement permitted rotation of the model for angle of attack changes, but all tests were run at 0° AOA. All models were tested in a vertical orientation in the center of the test section. Because of the rapid-start capability of the current IRT spray system, the models were not shielded during the initiation of the spray.



(a) NASA Glenn Icing Research Tunnel. Revised Loop with New Heat Exchanger Installed in 2000.

Figure 1. Facility and Model Description.

In preparing for a test, the temperature and airspeed in the test section and the air and water pressures on the spray manifolds were set. When these conditions had stabilized the spray nozzle valves were opened to initiate the spray. The spray was timed for the required duration, then turned off. The fan was brought to a full stop and the tunnel entered to record the ice shape. A thin heated plate with a cut-out of the model shape was inserted into the leading edge of the ice to melt a thin horizontal slice down to the clean model surface. A cardboard template was placed into this gap and an outline of the ice accretion traced. Tracings were taken at the vertical center of the tunnel (91 cm from the floor) and at 2.5 cm above the center. These



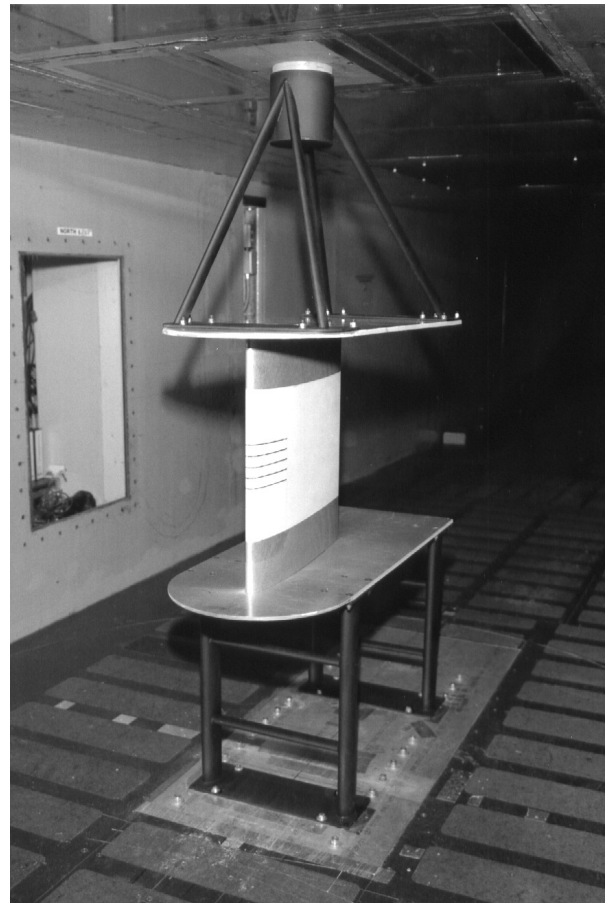
(b) 91.4-cm-Chord NACA 0012 Model in IRT Test Section.

Figure 1. (con't)

tracings were digitized and the x - y coordinates for each ice shape were recorded. The coordinates were normalized with respect to the model chord and plotted for comparison of ice shapes. In this paper, only centerline shapes will be presented.

Test Condition and Similarity Parameter Uncertainties

Tunnel and cloud conditions were recorded at 3-sec intervals over the duration of each test. Reported conditions are the time averages of these records. Estimates of the uncertainty in the reported average conditions were made by considering fluctuations of the values with time, possible instrument errors including calibration, uncertainties in tunnel calibration of MVD and LWC and observed differences in measurements from one location to another in the test section.



(c) 80.0-cm-Chord NACA 0012 Model in IRT Test Section. 53.3-cm-Chord Model Similar.

Figure 1. (concluded).

Total temperature was estimated to have an uncertainty of about $\pm 0.5^\circ\text{C}$, velocity, $\pm 1\text{m/s}$, liquid-water content, $\pm 10\%$ and median volume droplet diameter, $\pm 12\%$.

Using the methods of Coleman and Steele,¹² these estimated test-condition uncertainties led to an uncertainty of 17% in K_0 , 2% in β_0 , 12% in A_c , 15% in n , 1°C each in ϕ and θ , 2% in Re and 12% in We .

Scaling Approach

An objective of the SLD scaling tests described here was to determine if Appendix-C conditions could be used to simulate SLD conditions by applying established scaling methods. Because the Ruff scaling method⁴ for tunnels without pressure control has proven effective, it was chosen as the basis for the scaling applied. As noted in the Introduction, for scaling of either model size or test conditions, the

Ruff method requires that the similarity parameters K_0 , A_c , n and either ϕ or θ have the same respective values for both scale and reference conditions.

For this study, an MVD of 50 μm , the largest droplet size within the Appendix-C envelope, was chosen for all scale tests. Scale models were NACA 0012 airfoil sections of 91-, 80- and 53-cm chord. The scale temperature and LWC were found by matching ϕ and n to their respective reference values, and the scale velocity was determined using either the constant- We or average- V methods. With both model size and MVD selected for each scale test, the parameter K_0 (and therefore β_0) could not be matched. For all tests, however, the scale and reference β_0 's were found to differ by less than 10%. The normal practice with the Ruff method is to find scale icing time by simply equating scale and reference A_c . This insures the same relative quantity of ice accretion. However, because the scale and reference β_0 were not the same for these tests, the scale time was found by matching the product $A_c\beta_0$. The reference value chosen for A_c was the same as that of previous IRT scaling tests to facilitate comparisons with earlier results.

Results

Table II shows the matrix of scale tests performed. Only a representative sample of centerline shapes will be given here. There was no significant difference between the centerline shapes and those recorded 2.5 cm above center.

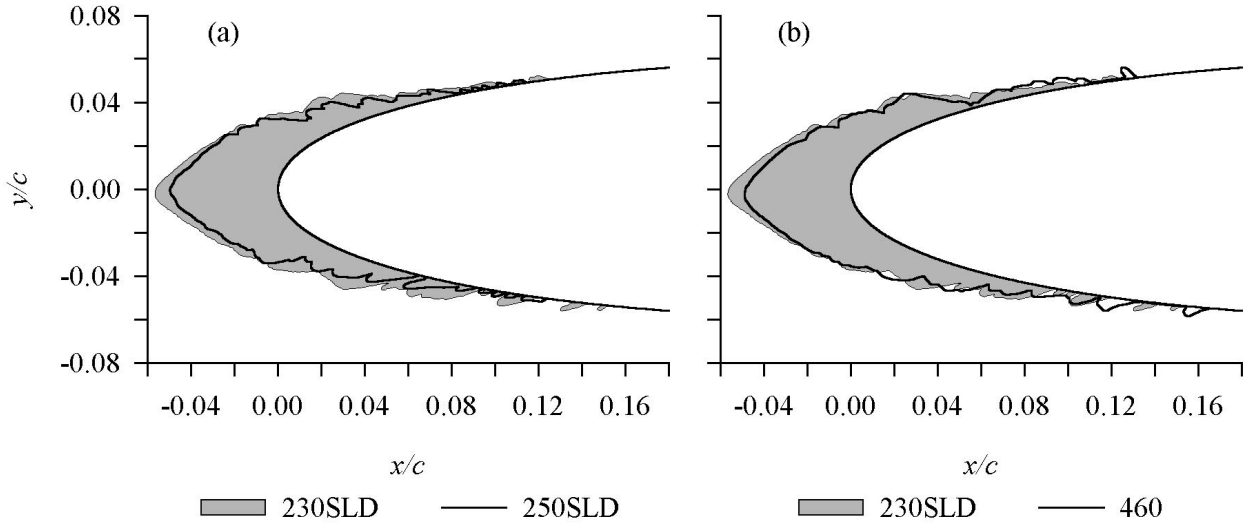
IRT Calibration Consistency

Rime tests made over a range of conditions while maintaining constant A_c and K_0 should produce identical ice shapes. Of particular interest for the present series of tests is whether the LWC calibration performed for SLD conditions is consistent with that of Appendix C. In figure 2, a 120- μm SLD shape is compared with a 70- μm SLD shape (fig. 2(a)) and with a 40- μm Appendix-C shape (fig. 2(b)). The 120- μm MVD shape for this figure is shown shaded, while the 70- and 40- μm shapes are each represented by a solid line. Below the figure are the average test conditions recorded during each test run and some of the corresponding similarity parameters of interest. Because the recorded conditions may have differed slightly from the planned set conditions for each

Table II. Matrix of SLD Scaling Tests
IRT, March 2001

Case	n	Reference $c = 91.4 \text{ cm}$			Average- V Scale $MVD = 50\mu\text{m}$			Constant- We Scale $MVD = 50\mu\text{m}$		
		V , kt	V , m/s	MVD , μm	c , 91.4 cm	c , 80.0 cm	c , 53.3 cm	c , 91.4 cm	c , 80.0 cm	c , 53.3 cm
130SLD	1.0	100	51.5	120					x	x
122SLD	0.6	100	51.5	175	x	x	x	x		
132SLD	0.6	100	51.5	120	x	x			x	
230SLD	1.0	150	77.2	120	x	x		x	x	
250SLD	1.0	150	77.2	70			x			x
222SLD	0.6	150	77.2	175	x	x	x	x	x	x
232SLD	0.6	150	77.2	120	x	x	x	x	x	x
242SLD	0.6	150	77.2	100	x	x	x	x	x	x
252SLD	0.6	150	77.2	70	x		x	x		x
225SLD	0.3	150	77.2	175	x	x	x			x
235SLD	0.3	150	77.2	120	x	x				x
245SLD	0.3	150	77.2	100	x	x				x

x indicates completed scale test



(a) Scaling from 120 to 70 μm MVD . Scale chord, 91.4 cm. (b) Scaling from 120 to 40 μm MVD . Scale chord, 53.3 cm.

Case	Date/Run	c , cm	t_{st} , $^{\circ}\text{C}$	V , m/s	MVD , μm	LWC , g/m^3	τ , min	K_0	β_0	A_c	n	b	ϕ , $^{\circ}\text{C}$	θ , $^{\circ}\text{C}$	Re , 10^4	We , 10^3
230SLD	3-12-01/6	91.4	-27.0	77.3	120	0.69	15.8	20.3	0.95	1.91	1.00	1.00	26.3	33.5	18.7	11.0
(a) 250SLD	3-9-01/7	91.4	-23.4	77.3	70	0.60	18.1	8.9	0.90	1.90	1.00	0.52	22.7	29.3	18.5	6.4
(b) 460	3-21-01/4	53.3	-26.1	66.8	40	1.00	7.3	5.8	0.86	1.89	1.00	0.59	25.5	33.0	9.6	2.7

Figure 2. IRT Calibration Consistency. Reference Tests for Three Rime Cases.

test, the similarity parameters intended to be maintained may not always have matched precisely.

The leading-edge thicknesses were slightly different, but otherwise the shapes were the same. Even the 120- μm feather size and location was simulated by the tests at the smaller drop sizes. The differences in leading-edge thickness were well within the uncertainty in LWC ; therefore, these results provide confidence that the IRT calibration was consistent for SLD and Appendix-C conditions.

Note also from figure 2 that although the stagnation collection efficiencies, β_0 , for each pair of tests did not match, they were close enough to produce good agreement of non-dimensional shapes and quantity of ice accreted. These results are consistent with the guideline⁹ that scale and reference β_0 values need only be within 10% for good match of ice shapes.

It is also worth noting that the ice shapes appeared to be independent of model span. The 91.4-cm-chord model was full span (1.8 m), while the 53.3-cm-chord model was partial span (0.6 m). This result indicates that flow around the ends of the partial-

span model did not seem to have affected the cloud at the center of the model.

Scaling from 70 μm to 50 μm

It was anticipated that scaling from 70 to 50 μm would be accomplished easily since this situation represents little change in MVD . Consequently, only two cases with a 70- μm reference MVD were included in the test matrix. Table II shows that for case 250SLD ($n = 1$) two scale tests at $MVD = 50 \mu\text{m}$ were performed for comparison with the reference, and for case 252SLD ($n = 0.6$) four scale tests were made.

For the case 250SLD scale tests anomalous ice shapes were obtained. These shapes were unusually smooth and appeared to be eroded. The cause is still under investigation, but these results were felt to be unreliable and the ice shapes will not be shown.

As expected, for case 252SLD excellent agreement was obtained between the scale and reference ice shapes for both methods of finding scale velocity and each scale model size. Because successful scal-

ing from larger MVD s will be presented below, the results of scaling from 70 μm will not be shown.

Scaling from 100 μm to 50 μm

Two 100- μm reference cases were included in the test matrix. Case 242SLD had a freezing fraction of 0.6 and case 245SLD had a freezing fraction of 0.3.

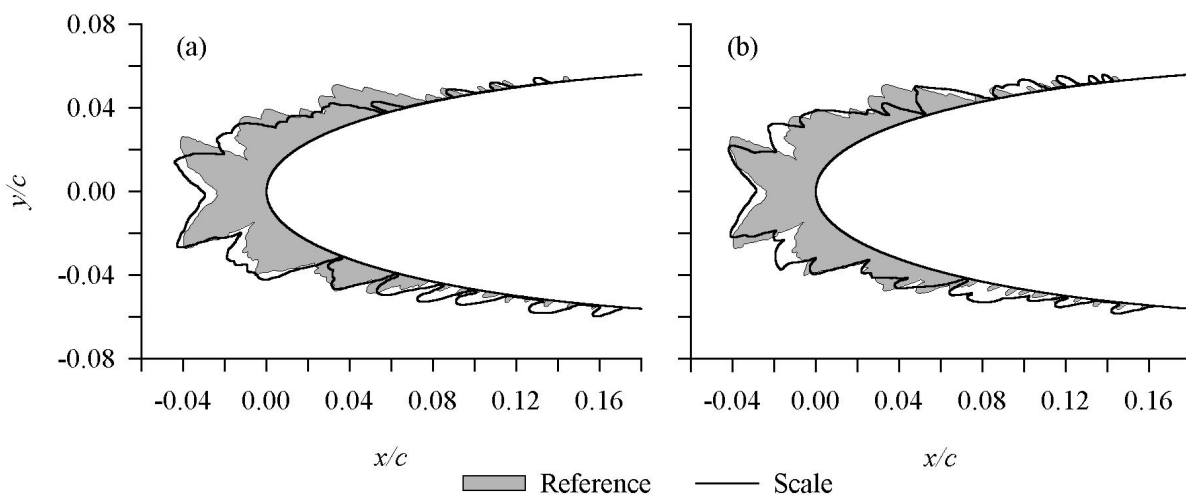
Six scale tests were performed for case 242SLD: each of the three scale models was tested using both average-velocity and constant- We methods to determine scale velocity. Figure 3(a) shows the scaling results using the average- V method with a scale chord of 53cm. Figure 3(b) gives the results for the constant- We method with the same scale model. The reference shape in this and subsequent figures is shown shaded while the scale shapes are each indicated by a solid line. Non-dimensional scale ice accretions matched the reference in quantity of ice, in shape and location of horns and in size and approximate locations of feather formations. Even the large feather structures just aft of the main ice horns were simulated reasonably well by the scale tests. There was no significant difference in results whether the average-velocity method or constant- We method was used to choose scale velocity.

The ice shapes at a freezing fraction of 0.3 (Case 245SLD) differed in appearance from those obtained at 0.6, but the scale tests using either scaling method matched the reference in non-dimensional size and shape. These results will not be shown.

Scaling from 120 μm to 50 μm

Five cases had a reference MVD of 120 μm . Two of these had a reference V of 100 kt (130SLD, with $n = 1.0$ and 132SLD, with $n = 0.6$) and three with a reference V of 150 kt (230SLD, with $n = 1.0$, 232SLD, with $n = 0.6$ and 235SLD, with $n = 0.3$). The scale tests for the rime cases, 130SLD and 230SLD, produced ice shapes that looked eroded, similar to those mentioned above for the 70- to 50- μm scaling tests, and these will not be discussed further.

Three scaling tests were performed for case 132SLD. Two of these used the average- V method (scale model chords of 91.4 and 80 cm) and one, the constant- We (80.0-cm scale model chord). Within typical repeatability of ice shapes, all scale tests simulated the reference accretion in terms of the non-dimensional size, horn location, horn angle and feather size and density. There was a tendency for the non-dimensional scale horns to be slightly larger



(a) Average- V method. Scale chord, 53.3 cm.

(b) Constant- We method. Scale chord, 53.3 cm.

Method	Date/Run	c , cm	t_{st} , °C	V , m/s	MVD , μm	LWC , g/m^3	τ , min	K_0	β_0	A_c	n	b	ϕ , °C	θ , °C	Re , 10^4	We , 10^3
Reference	3-9-01/1	91.4	-15.7	77.5	100	0.68	16.0	15.3	0.93	1.91	0.60	0.61	15.0	20.0	17.6	9.2
(a) Avg- V	3-19-01/9	53.3	-16.8	126.2	50	0.56	7.0	12.1	0.92	1.92	0.60	0.50	14.9	16.4	16.0	12.3
(b) C- We	3-20-01/4	53.3	-16.3	109.2	50	0.67	6.8	11.1	0.91	1.92	0.60	0.55	14.9	17.8	14.0	9.2

Figure 3. Scaling from 100 to 50 μm MVD with freezing fraction of 0.6. Case 242SLD. Reference chord, 91.4 cm.

than the reference, but this difference was not judged significant. Scale results were independent of both scale model size and method of determining scale velocity.

Scale tests for case 232SLD, with a freezing fraction of 0.6, included both methods of finding scale velocity applied to all three model sizes. All of these scaling tests produced at least fair agreement with the reference shape. Two examples are shown in figure 4, which compares the 53-cm scale model results with reference shapes.

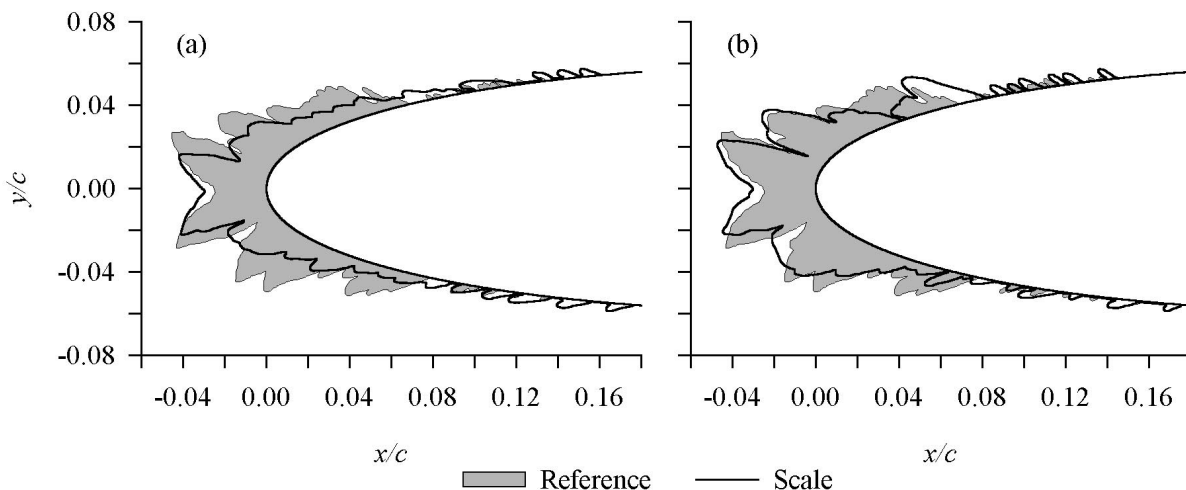
The average- V scaling for this case, shown in figure 4(a), gave the poorest match of non-dimensional shape and quantity of the three model sizes tested with this method, while the constant- We results in 4(b) illustrate the best of the three model sizes. For this latter test, the reference ice shape horns were simulated well by the scale result, as were the sizes and approximate spacing of feather structures aft of the main shape. It was concluded that differences between the methods were only random in nature, since each gave excellent scaling results at one of the model sizes. Additional tests of this case are needed, however, to confirm this conclusion.

Case 235SLD had a freezing fraction of 0.3. Two scaling tests were made with the average- V method, and results of these are given in figure 5. Figure 5(a) shows scale ice shapes for a scale model of 91.4-cm chord, and 5(b) for a scale model of 80.0-cm chord. Both scale tests produced an excellent simulation of the reference ice shape, including the horn size, location and angle and feather sizes and spacing. The constant- We method with a model chord of 53.3 cm was also tested with similarly excellent results. This shape is not illustrated.

Scaling from 175 μm to 50 μm

Three 175- μm -reference cases were included in the study. The non-dimensional scale ice shapes for cases 122SLD and 222SLD were markedly smaller than the reference shapes and for some tests appeared to be eroded. Additional study is required to understand these results.

Case 225SLD had a 150-kt reference velocity and a freezing fraction of 0.3. Average- V scaling was applied for each of the three model sizes, and constant- We was used for the 53.3-cm model. Figure 6 shows typical scaling results. Figure 6(a) shows results for

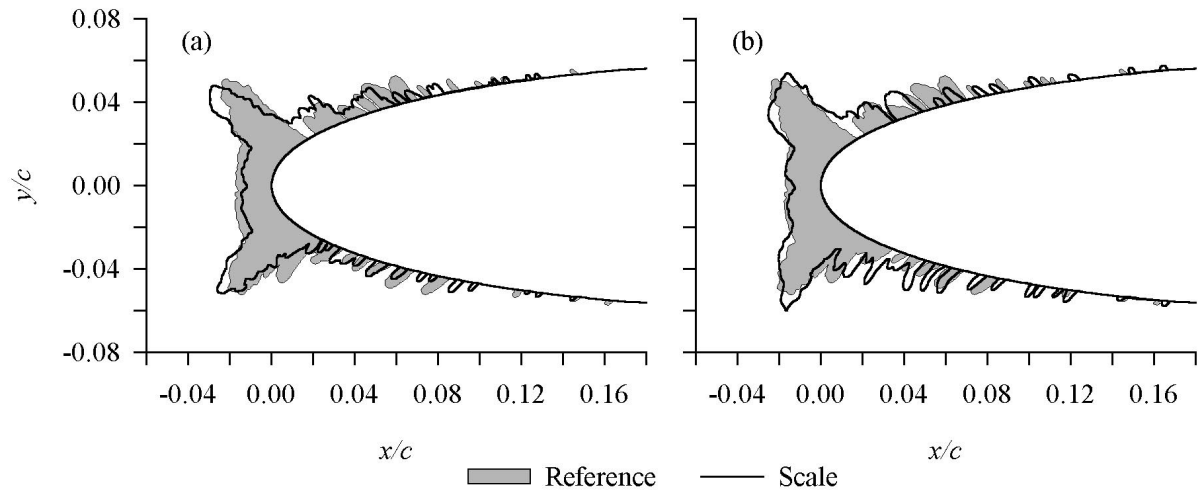


(a) Average- V method. Scale chord, 53.3 cm.

(b) Constant- We method. Scale chord, 53.3 cm.

Method	Date/Run	c , cm	t_{st} , °C	V , m/s	MVD , μm	LWC , g/m^3	τ , min	K_0	β_0	A_c	n	b	ϕ , °C	θ , °C	Re , 10^4	We , 10^3
Reference	3-9-01/2	91.4	-16.0	77.5	120	0.69	15.8	20.2	0.95	1.91	0.60	0.64	15.3	20.5	17.7	11.1
(a) Avg- V	3-19-01/8	53.3	-17.3	131.5	50	0.55	6.9	12.3	0.92	1.93	0.60	0.50	15.2	16.3	16.6	13.2
(b) C- We	3-19-01/10	53.3	-16.8	119.3	50	0.62	6.8	11.8	0.92	1.96	0.60	0.54	15.2	17.3	15.3	11.0

Figure 4. Scaling from 120 to 50 μm MVD with freezing fraction of 0.6. Case 232SLD. Reference chord, 91.4 cm.

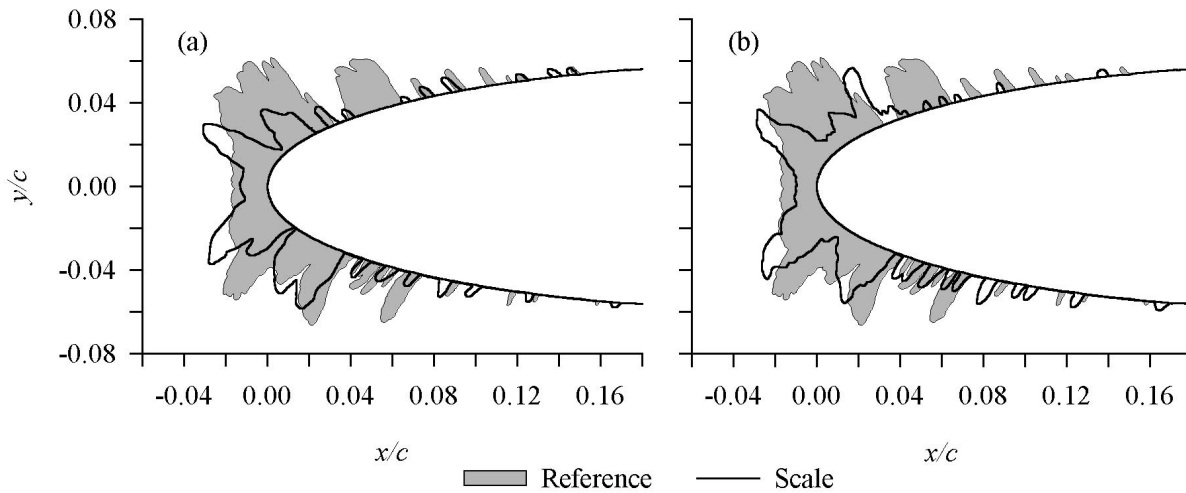


(a) Scale chord, 91.4 cm.

(b) Scale chord, 80.0 cm.

Method	Date/Run	c , cm	t_{st} , °C	V , m/s	MVD , μm	LWC , g/m^3	τ , min	K_0	β_0	A_c	n	b	ϕ , °C	θ , °C	Re , 10^4	We , 10^3
Reference	3-7-01/4	91.4	-8.4	77.0	120	0.69	15.8	20.2	0.95	1.90	0.30	0.63	7.7	10.2	16.7	10.9
(a) Avg- V	3-7-01/1	91.4	-8.9	98.7	50	0.58	14.7	6.2	0.86	1.90	0.30	0.56	7.7	8.9	21.0	7.5
(b) Avg- V	3-13-01/1	80.0	-8.9	104.3	50	0.56	13.5	7.3	0.88	2.05	0.30	0.54	7.6	8.5	18.9	8.4

Figure 5. Scaling from 120 to 50 μm MVD with freezing fraction of 0.3. Case 235SLD. Reference chord, 91.4 cm.



(a) Average- V method. Scale chord, 53.3 cm.

(b) Constant- We method. Scale chord, 53.3 cm.

Method	Date/Run	c , cm	t_{st} , °C	V , m/s	MVD , μm	LWC , g/m^3	τ , min	K_0	β_0	A_c	n	b	ϕ , °C	θ , °C	Re , 10^4	We , 10^3
Reference	3-7-01/5	91.4	-10.4	77.4	175	0.99	11.0	35.6	0.97	1.91	0.30	0.93	9.7	13.0	17.0	16.1
(a) Avg- V	3-19-01/2	53.3	-12.1	143.2	50	0.58	6.2	13.1	0.92	1.99	0.30	0.56	9.6	7.9	17.1	15.8
(b) C- We	3-19-01/1	53.3	-11.1	136.5	50	0.59	6.2	12.8	0.92	1.95	0.28	0.56	8.9	7.5	16.4	14.4

Figure 6. Scaling from 175 to 50 μm MVD ; Freezing Fraction, 0.3; Reference chord, 91.4 cm. Case 225SLD.

the average- V method and 6(b) for the constant- We method for the 53.3-cm model.

Both methods of finding scale velocity produced scale shapes that were non-dimensionally smaller than the reference, exhibited horn structures different from the reference and failed to simulate the large feather formations aft of the main reference ice shape. Splashing of impinging droplets might redistribute water on the surface to create significantly different shapes, and, based on the Mundo, Sommerfeld and Tropea³ K factor for splashing, splashing would tend to be more significant in the reference tests. However, this explanation is inconsistent with the larger quantity of ice accreted for the 175- μm MVD . Significant differences in the droplet size distribution for the 175- and 50- μm MVD clouds might also account for the shape disparities, but the distributions reported by Ide and Oldenburg¹⁰ for the IRT do not reveal such differences. Testing is planned to study scaling from 175- μm MVD clouds further.

Conclusions

- Scaling to Appendix-C conditions from SLD was accomplished successfully for reference MVD 's up to 120 μm . This result suggests that either splashing does not significantly affect ice shapes for drop sizes up to 120 μm , or the two scaling methods used in this study already adequately describe splashing phenomena.
- Scaling from 175 μm MVD to 50 μm was not successful. Because the reference ice shapes were non-dimensionally much larger than the scale, splashing does not seem to be the likely explanation for the dissimilarity. Additional testing is required to identify the cause of the shape differences between scale and reference.
- No evidence of end effects was observed in tests with the partial-span models. Full-span and partial-span models gave consistent ice shapes.
- Both the constant- We and average- V methods of determining scale velocity gave equivalent scaling results.

References

- ¹ Bilanin, Alan J. and Anderson, David N., "Ice Accretion with Varying Surface Tension," AIAA-95-0538 and NASA TM-106826, January 1995.
- ² Wright, William B., "Computational Simulation of Large Droplet Icing," *Proceedings of the FAA International Conference on Aircraft Inflight Icing* vol. II, DOT/FAA/AR-96/81,II, August 1996, pp 545-555.
- ³ Mundo, C., Sommerfeld, M. and Tropea, C., "Droplet-Wall Collisions: Experimental Studies of the Deformation and Breakup Process," *Int. J. Multiphase Flow*, vol. 21, no. 2, 1995, pp. 151-173.
- ⁴ Ruff, Gary A., "Analysis and Verification of the Icing Scaling Equations," AEDC-TR-85-30, vol. 1 (rev), March 1986.
- ⁵ Messinger, B.L., "Equilibrium Temperature of an Unheated Icing Surface as a Function of Airspeed," *J. Aeron. Sci.*, vol. 20 no. 1, January 1953, pp. 29-42.
- ⁶ Anderson, David N. and Ruff, Gary A., "Evaluation of Methods to Select Scale Velocities in Icing Scaling Tests," AIAA-99-0244, January 1999.
- ⁷ Anderson, David N., "Effect of Velocity in Icing Scaling Tests," AIAA-2000-0236, January 2000.
- ⁸ Irvine, Thomas B. and Anderson, David N., "The NASA Lewis Icing Research Tunnel Testing and Data Acquisition," *Proceedings of the FAA International Conference on Aircraft Inflight Icing* vol. II, DOT/FAA/AR-96/81,II, August 1996, pp. 469-483.
- ⁹ Anderson, David N., "Acceptable Tolerances for Matching Icing Similarity Parameters in Scaling Applications," AIAA-2001-0832, January 2001.
- ¹⁰ Ide, Robert F. and Oldenburg, John R., "Icing Cloud Calibration of the NASA Glenn Icing Research Tunnel," AIAA-2001-0234, January 2001.
- ¹¹ Irvine, Thomas B., Kevdizija, Susan L., Sheldon, David W. and Spera, David A., "Overview of the Icing and Flow Quality Improvements Program for the NASA Glenn Icing Research Tunnel," AIAA-2001-0229, January 2001.
- ¹² Coleman, Hugh W. and Steele, W. Glenn, "Experimentation and Uncertainty Analysis for Engineers," John Wiley & Sons, 1989.

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13. ABSTRACT (Maximum 200 words) Proposed changes to aircraft icing certification rules are being considered by European, Canadian, and American regulatory agencies to include operation in super-cooled large droplet conditions (SLD). This paper reports results of an experimental study in the NASA Glenn Icing Research Tunnel (IRT) to evaluate how well scaling methods developed for Appendix C conditions might apply to SLD conditions. Until now, scaling studies have been confined to the FAA FAR-25 Appendix C envelope of atmospheric cloud conditions. Tests were made in which it was attempted to scale to a droplet <i>MVD</i> of 50µm from clouds having droplet <i>MVDs</i> of 175, 120, 100, and 70µm. Scaling was based on the Ruff method with scale velocities found either by maintaining constant Weber number or by using the average of the velocities obtained by maintaining constant Weber number and constant Reynolds number. Models were unswept NACA 0012 wing sections. The reference model had a chord of 91.4 cm. Scale models had chords of 91.4, 80.0, and 53.3 cm. Tests were conducted with reference airspeeds of 100 and 150 kt (52 and 77 m/s) and with freezing fractions of 1.0, 0.6, and 0.3. It was demonstrated that the scaled 50-µm cloud simulated well the non-dimensional ice shapes accreted in clouds with <i>MVD</i> 's of 120µm or less.				
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